Abstract:

The invention relates to a compressor, particularly a high-pressure compressor, of a gas turbine, particularly of an aircraft engine. The compressor comprises at least one rotor and a number of blades (11, 12), which are assigned to the or to each rotor and which rotate together with the respective rotor. Each blade (11, 12) is delimited, in essence, by a flow entry edge or leading edge (16), a flow exit edge or trailing edge (17), and by a blade surface (20), which extends between the leading edge (16) and the trailing edge (17) while forming a suction side (18) and a pressure side. According to the invention, the leading edges (16) of the blades (11, 12) are slanted at a sweep angle that changes with the height of the respective blade (11, 12) in such a manner that the leading edges (11) comprise, in a radially external area (23) of the same, at least one forward sweep angle, a backward sweep angle or zero-sweep angle following in a radially external manner, and a forward sweep angle following, in a radially external manner, the backward sweep angle or the zero-sweep angle.